

ABSTRACT OF THE DISCLOSURE

The invention relates to an annular combustor (13) for a gas turbine (10), into which combustor (13) burners (14, 15) open on an inlet side, and which combustor (13) extends in the axial direction from the inlet side to an outlet side (33) and is lined on the insides with cooled liner segments (16, 17) for protection from the hot gases.

In such a combustor, increased mechanical stability and flexibility in design and also simplification in manufacture and fitting are achieved by the liner segments (16, 17) being subdivided in the axial direction into a plurality of parts (16, 17) arranged one behind the other.

(Fig. 1)